

Compressible Flow - TME085

Lecture 10

Niklas Andersson

Chalmers University of Technology
Department of Mechanical Engineering
Division of Fluid Mechanics
Gothenburg, Sweden

`niklas.andersson@chalmers.se`





Chapter 7 - Unsteady Wave Motion

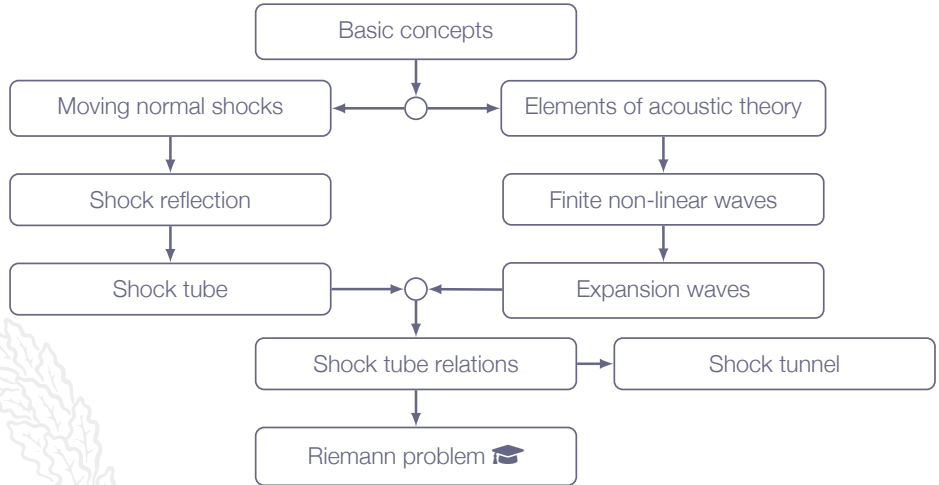


Learning Outcomes

- 3 **Describe** typical engineering flow situations in which compressibility effects are more or less predominant (e.g. Mach number regimes for steady-state flows)
- 4 **Present** at least two different formulations of the governing equations for compressible flows and **explain** what basic conservation principles they are based on
- 8 **Derive** (marked) and **apply** (all) of the presented mathematical formulae for classical gas dynamics
 - a 1D isentropic flow*
 - b normal shocks*
 - j unsteady waves and discontinuities in 1D
 - k basic acoustics
- 9 **Solve** engineering problems involving the above-mentioned phenomena (8a-8k)
- 11 **Explain** how the equations for aero-acoustics and classical acoustics are derived as limiting cases of the compressible flow equations

moving normal shocks - frame of reference seems to be the key here?!

Roadmap - Unsteady Wave Motion



Motivation

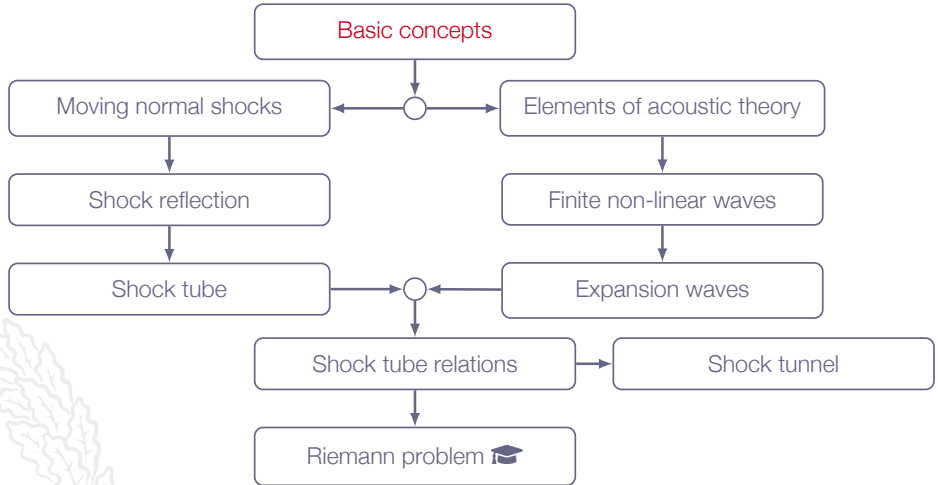
Most practical flows are unsteady

Traveling waves appears in many real-life situations and is an important topic within compressible flows

We will study unsteady flows in one dimension in order to reduce complexity and focus on the physical effects introduced by the unsteadiness

Throughout this section, we will study an application called the shock tube, which is a rather rare application but it lets us study unsteady waves in one dimension and it includes all physical principles introduced in chapter 7

Roadmap - Unsteady Wave Motion



Unsteady Wave Motion - Example #1

Object moving with supersonic speed through the air

observer moving with the bullet

steady-state flow

the detached shock wave is

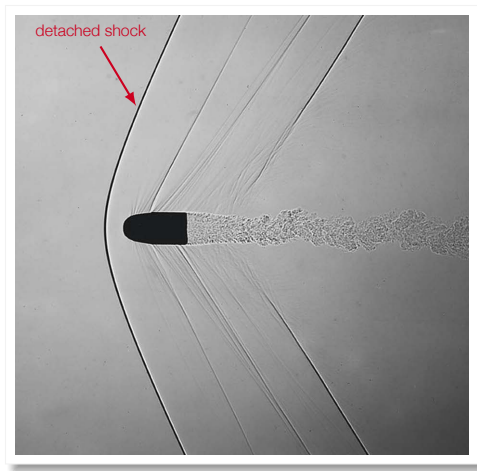
stationary

observer at rest

unsteady flow

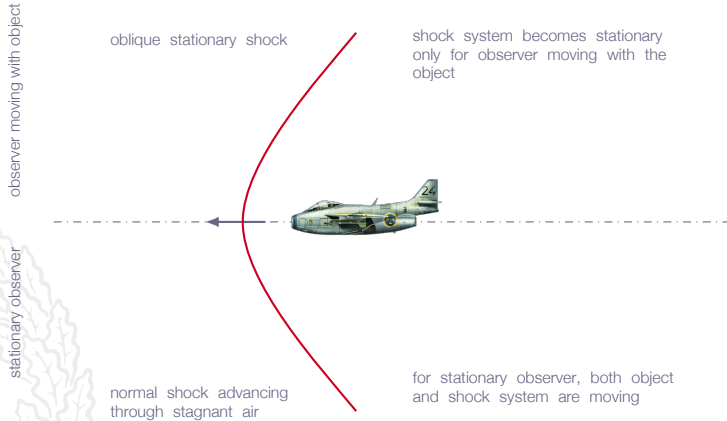
detached **shock wave moves**

through the air (to the left)



Unsteady Wave Motion - Example #2

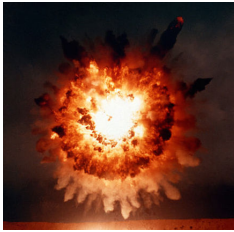
Object moving with supersonic speed through the air



Unsteady Wave Motion - Example #3

Shock wave from explosion

MID-AIR EXPLOSION



GROUND EXPLOSION

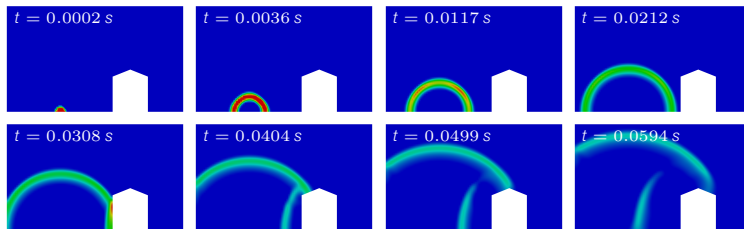


For observer at rest with respect to the surrounding air:

1. the flow is **unsteady**
2. the **shock wave moves** through the air

Unsteady Wave Motion - Example #4

Shock wave from explosion



normal shock moving spherically outwards:

shock **strength decreases** with radius

shock **speed decreases** with radius

Unsteady Wave Motion

inertial frames!

Physical laws are the same for both frame of references

Shock characteristics are the same for both observers (shape, strength, etc)

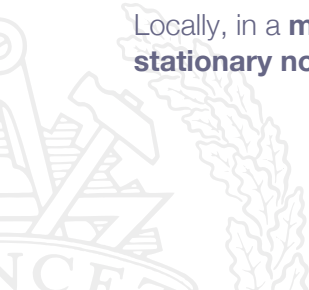
Recall - the Hugoniot relation does not include velocities, only static thermodynamic quantities that are independent of reference frame

Unsteady Wave Motion

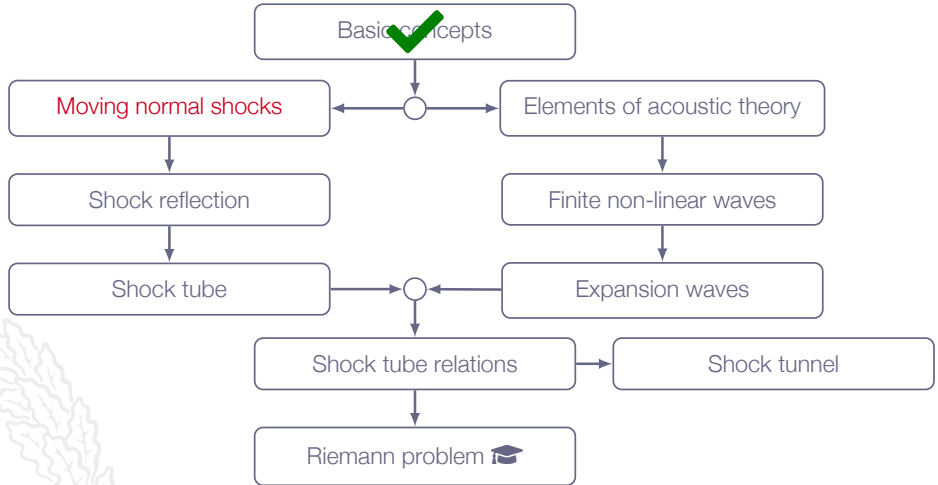
Is there a connection with stationary shock waves?

Answer: Yes!

Locally, in a **moving frame of reference**, the shock may be viewed as a **stationary normal shock**



Roadmap - Unsteady Wave Motion



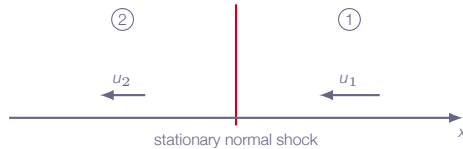
Chapter 7.2

Moving Normal Shock Waves



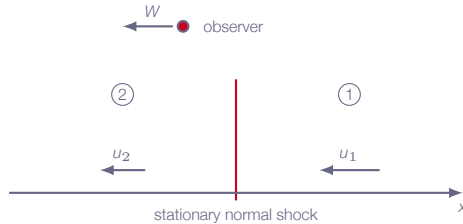
Moving Normal Shock Waves

Chapter 3: stationary normal shock



- $u_1 > a_1$ (supersonic flow)
- $u_2 < a_2$ (subsonic flow)
- $p_2 > p_1$ (sudden compression)
- $s_2 > s_1$ (shock loss)

Moving Normal Shock Waves



Introduce **observer moving to the left** with speed W
if W is **constant** the observer is still in an **inertial system**
(all physical laws are unchanged)

The observer sees a **normal shock moving to the right** with speed W
gas velocity ahead of shock: $u'_1 = W - u_1$
gas velocity behind shock: $u'_2 = W - u_2$

Moving Normal Shock Waves

Now, let $W = u_1 \Rightarrow$

$$u'_1 = 0$$

$$u'_2 = u_1 - u_2 > 0$$

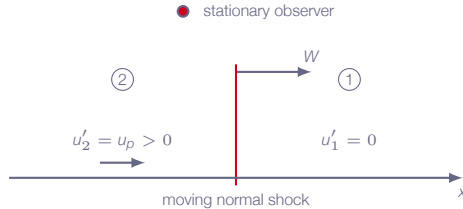
The observer now sees the shock traveling to the right with speed $W = u_1$ into a stagnant gas, leaving a compressed gas ($p_2 > p_1$) with velocity $u'_2 > 0$ behind it

Introducing u_p :

$$u_p = u'_2 = u_1 - u_2$$

Moving Normal Shock Waves

Analogy:



Case 1

stationary normal shock

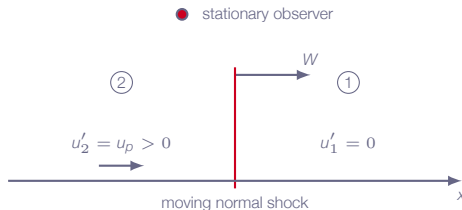
observer moving with velocity W

Case 2

normal shock moving with velocity W

stationary observer

Moving Normal Shock Waves - Governing Equations



For stationary normal shocks we have:

$$\begin{aligned}\rho_1 u_1 &= \rho_2 u_2 \\ \rho_1 u_1^2 + p_1 &= \rho_2 u_2^2 + p_2 \\ h_1 + \frac{1}{2} u_1^2 &= h_2 + \frac{1}{2} u_2^2\end{aligned}$$

With ($u_1 = W$) and ($u_2 = W - u_p$) we get:

$$\begin{aligned}\rho_1 W &= \rho_2 (W - u_p) \\ \rho_1 W^2 + p_1 &= \rho_2 (W - u_p)^2 + p_2 \\ h_1 + \frac{1}{2} W^2 &= h_2 + \frac{1}{2} (W - u_p)^2\end{aligned}$$

Moving Normal Shock Waves - Relations

Starting from the governing equations

$$\rho_1 W = \rho_2 (W - u_p)$$

$$\rho_1 W^2 + p_1 = \rho_2 (W - u_p)^2 + p_2$$

$$h_1 + \frac{1}{2} W^2 = h_2 + \frac{1}{2} (W - u_p)^2$$

and using $h = e + \frac{p}{\rho}$

it is possible to show that

$$e_2 - e_1 = \frac{p_1 + p_2}{2} \left(\frac{1}{\rho_1} + \frac{1}{\rho_2} \right)$$

Moving Normal Shock Waves - Relations

$$e_2 - e_1 = \frac{p_1 + p_2}{2} \left(\frac{1}{\rho_1} + \frac{1}{\rho_2} \right)$$

same Hugoniot equation as for stationary normal shock

This means that we will have same shock strength, *i.e.* same discontinuities in density, velocity, pressure, etc

Moving Normal Shock Waves - Relations

Starting from the Hugoniot equation one can show that

$$\frac{\rho_2}{\rho_1} = \frac{1 + \frac{\gamma + 1}{\gamma - 1} \left(\frac{p_2}{p_1} \right)}{\frac{\gamma + 1}{\gamma - 1} + \frac{p_2}{p_1}}$$

and

$$\frac{T_2}{T_1} = \frac{p_2}{p_1} \left[\frac{\frac{\gamma + 1}{\gamma - 1} + \frac{p_2}{p_1}}{1 + \frac{\gamma + 1}{\gamma - 1} \left(\frac{p_2}{p_1} \right)} \right]$$

Moving Normal Shock Waves - Relations

For calorically perfect gas and stationary normal shock:

$$\frac{\rho_2}{\rho_1} = 1 + \frac{2\gamma}{\gamma + 1}(M_s^2 - 1)$$

same as eq. (3.57) in Anderson with $M_1 = M_s$

where

$$M_s = \frac{W}{a_1}$$

M_s is simply the speed of the shock, traveling into the stagnant gas, normalized by the speed of sound in the gas ahead of the shock

Note!

$M_s > 1$, otherwise there is no shock!

shocks always moves faster than sound - no warning before it hits you ☺

Moving Normal Shock Waves - Relations

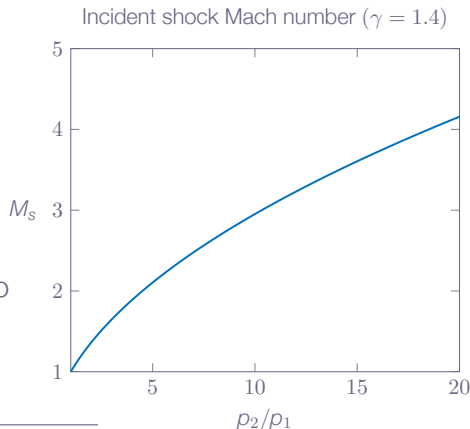
$$\frac{p_2}{p_1} = 1 + \frac{2\gamma}{\gamma + 1}(M_s^2 - 1)$$

Re-arrange $\Rightarrow$

$$M_s = \sqrt{\frac{\gamma + 1}{2\gamma} \left(\frac{p_2}{p_1} - 1 \right) + 1}$$

shock speed directly linked to pressure ratio

$$M_s = \frac{W}{a_1} \Rightarrow W = a_1 M_s = a_1 \sqrt{\frac{\gamma + 1}{2\gamma} \left(\frac{p_2}{p_1} - 1 \right) + 1}$$



Moving Normal Shock Waves - Induced Flow Velocity

From the continuity equation we get:

$$u_p = W \left(1 - \frac{\rho_1}{\rho_2} \right) > 0$$

After some derivation we obtain:

$$u_p = \frac{a_1}{\gamma} \left(\frac{p_2}{p_1} - 1 \right) \left[\frac{\frac{2\gamma}{\gamma+1}}{\frac{p_2}{p_1} + \frac{\gamma-1}{\gamma+1}} \right]^{1/2}$$

Moving Normal Shock Waves - Induced Flow Mach Number

$$M_p = \frac{u_p}{a_2} = \frac{u_p}{a_1} \frac{a_1}{a_2} = \frac{u_p}{a_1} \sqrt{\frac{T_1}{T_2}}$$

inserting u_p/a_1 and T_1/T_2 from relations on previous slides we get:

$$M_p = \frac{1}{\gamma} \left(\frac{p_2}{p_1} - 1 \right) \left[\frac{\frac{2\gamma}{\gamma+1}}{\frac{\gamma-1}{\gamma+1} + \frac{p_2}{p_1}} \right]^{1/2} \left[\frac{1 + \left(\frac{\gamma+1}{\gamma-1} \right) \left(\frac{p_2}{p_1} \right)}{\left(\frac{\gamma+1}{\gamma-1} \right) \left(\frac{p_2}{p_1} \right) + \left(\frac{p_2}{p_1} \right)^2} \right]^{1/2}$$

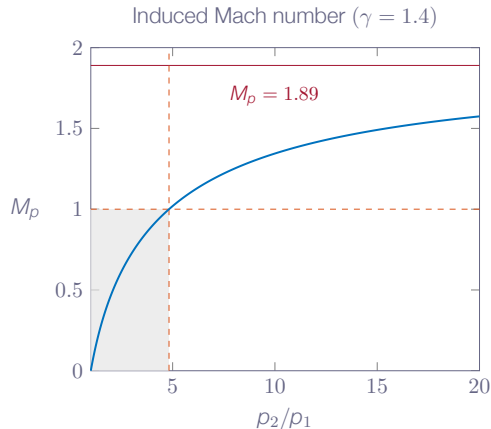
Moving Normal Shock Waves - Induced Flow Mach Number

Note!

$$\lim_{\frac{p_2}{p_1} \rightarrow \infty} M_p \rightarrow \sqrt{\frac{2}{\gamma(\gamma - 1)}}$$

for air ($\gamma = 1.4$)

$$\lim_{\frac{p_2}{p_1} \rightarrow \infty} M_p \rightarrow 1.89$$



Moving Normal Shock Waves - Example

Moving normal shock with $p_2/p_1 = 10$

$(p_1 = 1.0 \text{ bar}, T_1 = 300 \text{ K}, \gamma = 1.4)$

$\Rightarrow M_s = 2.95$ and $W = 1024.2 \text{ m/s}$

The shock is advancing with almost **three times** the speed of sound!

Behind the shock the induced velocity is $u_p = 756.2 \text{ m/s} \Rightarrow$ supersonic flow
($a_2 = 562.1 \text{ m/s}$)

May be calculated by formulas 7.13, 7.16, 7.10, 7.11 or by using Table A.2 for stationary normal shock ($u_1 = W, u_2 = W - u_p$)

Moving Normal Shock Waves - Total Enthalpy

Note! $h_{o1} \neq h_{o2}$

constant total enthalpy is **only valid for stationary shocks!**

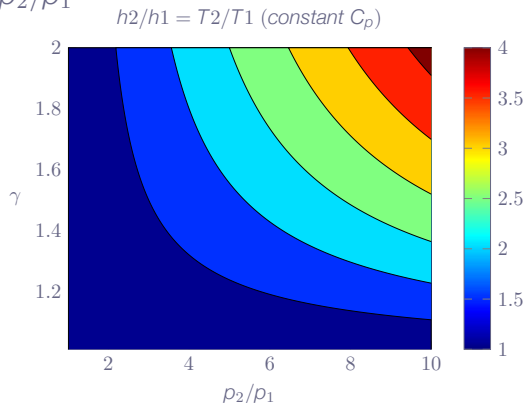
shock is uniquely defined by pressure ratio p_2/p_1

$$u_1 = 0$$

$$h_{o1} = h_1 + \frac{1}{2}u_1^2 = h_1$$

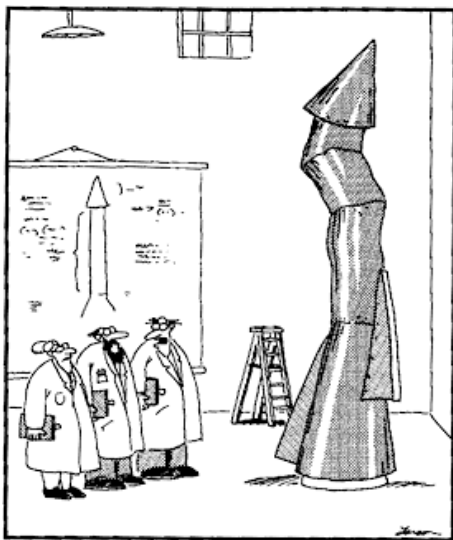
$$h_{o2} = h_2 + \frac{1}{2}u_2^2$$

$$h_2 > h_1 \Rightarrow h_{o2} > h_{o1}$$



Moving Normal Shock Waves - Total Enthalpy

Gas/Vapor	Ratio of specific heats (γ)	Gas constant R
Acetylene	1.23	319
Air (standard)	1.40	287
Ammonia	1.31	530
Argon	1.67	208
Benzene	1.12	100
Butane	1.09	143
Carbon Dioxide	1.29	189
Carbon Disulphide	1.21	120
Carbon Monoxide	1.40	297
Chlorine	1.34	120
Ethane	1.19	276
Ethylene	1.24	296
Helium	1.67	2080
Hydrogen	1.41	4120
Hydrogen chloride	1.41	230
Methane	1.30	518
Natural Gas (Methane)	1.27	500
Nitric oxide	1.39	277
Nitrogen	1.40	297
Nitrous oxide	1.27	180
Oxygen	1.40	260
Propane	1.13	189
Steam (water)	1.32	462
Sulphur dioxide	1.29	130



"It's time we face reality, my friend. ... We're not exactly rocket scientists."